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#### ABSTRACTS FROM THE SCIENTIFIC AND TECHNICAL PRESS.

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(Prepared by R.T.P.3.)

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The Superposition of Undamped Two-Dimensional Pressure Waves of Large Amplitude. (H. Pfriem, Akustische Zeitschrift, Vol. 7, No. 2, March, 1942, pp. 56-65.) (122/1 Germany.)

The author considers the single large amplitude pressure wave as made up of elementary waves following each other in succession, each wave travelling at the particular speed corresponding to the condition existing in the wake of its predecessor. In the case of an originally homogeneous atmosphere, each elementary wave will travel at a constant speed which however will differ from element to element. If, however, two elementary waves overlap, due to two pressure pulses travelling in opposite direction along the same path, the gas particle subjected to the simultaneous action of two elementary waves will assume a resultant state which will determine the new speed of propagation. If  $a_1$ ,  $u_1$  represent the sonic state and velocity at given particles when subjected only to

the action of elementary wave I travelling in the +x direction (i.e., direction of increasing particle velocity) and  $a_2$ ,  $u_2$  represent similar effects of wave 2 travelling in the opposite direction, the author shows the new velocities of I and 2 on meeting as given by:—

$$+(a+u) = \left\{ \frac{(\gamma+1)}{(\gamma-1)} \right\} a_1 - \left\{ \frac{(3-\gamma)}{(\gamma-1)} \right\} a_2 - a_0 + u_0 . \tag{1}$$

$$-(a-u) = -\left\{\frac{(\gamma+1)}{(\gamma-1)}\right\} a_2 - \left\{\frac{(3-\gamma)}{(\gamma-1)}\right\} a_1 - a_0 - u_0. \tag{2}$$

where  $a_0$  and  $u_0$  refer to conditions in the undisturbed atmosphere. The problem of superposition of large amplitude waves is therefore solved provided we can determine the time and place of meeting of the elementary waves of known sonic state. The author shows that this problem reduces itself to the solution of the two simultaneous partial differential equations:—

$$\frac{\partial a_1}{\partial t} + (a+u)\frac{\partial a_1}{\partial x} = 0 . (3)$$

$$\frac{\partial a_2}{\partial t} - (a - u) \frac{\partial a_2}{\partial x} = 0 \qquad . \tag{4}$$

The general solution of these equations has been given by Riemann but is exceedingly complex. Reasonably simple expressions for x and t are however obtained, if the gas is supposed to be monoatomic. In this case we obtain:—

$$\begin{aligned} x - (a + u) \ t &= \frac{\eth}{\eth a_1} \left[ \frac{F \left( a_1 \right) + G \left( a_2 \right)}{a} \right] \\ x + (a - u) \ t &= \frac{\eth}{\eth a_2} \left[ \frac{F \left( a_1 \right) + G \left( a_2 \right)}{a} \right] \end{aligned}$$

where  $F\left(a_1\right)$  and  $G\left(a_2\right)$  are functions depending on the shapes of the two large amplitude disturbances and are considered as given. Knowing the time and place of meeting of two elementary waves, the shape of the pressure waves in the zone of superposition can be determined. It appears that the mutual distortion is strictly limited to this field of superposition and that the waves, after leaving this zone and re-entering the original homogeneous atmosphere, have exactly the same shape as if they had progressed separately through the homogeneous atmosphere. They thus differ from their original shape only by the general steepening of the front associated with the normal propagation of large amplitude waves. It is interesting to note that in the special case of a gas with  $\gamma=3$ , the distortion of the waves in the field of superposition is eliminated. This follows from (1) and (2) where, on making the necessary substitutions, we find:—

$$a + u = 2a_1 - a_0 + u_0$$

and

$$a - u = 2a_2 - a_0 - u_0$$

Inserting these values in (3) and (4), we see that the simultaneous differential equations now become mutually independent, each representing a progressive wave travelling at a speed depending only on its proper sonic condition.

On the Question of Using Unshrouded Blades in Gas Turbines. (I. Shevyakov, Sovietskoye, Kotloturbostroynie, No. 1, Jan., 1940, pp. 30-32.) (122/2 U.S.S.R.)

In impulse turbines using steam, it is usual to surround the outer blade tips with a ring or shroud. This reduces leakage losses and also strengthens the blade assembly, but obviously raises the blade temperature since the cooling effect due to the rotor is reduced. Efficient blade cooling is the chief limiting factor in gas turbines and the question thus naturally arises whether in their

case the possibility of utilising a higher gas temperature in the absence of a shroud outweighs the drop in power due to increased leakage. For this purpose the author has carried out model experiments on a rotor sector containing six blades and subjected to the impact of air at about 200 m./sec., the reaction being statically balanced. The blades were 17 mm. in height, the corresponding nozzle exit diameter being 15 mm.

Measurements were made both with shrouded and unshrouded blades, intermediate readings being also taken with the shroud placed at various distances from the blade tips and thus providing radial clearances varying from 0 to 3 mm. Over this range, the result can be represented by the equation

$$\lambda = \frac{.04}{(.16 + \sigma)} + .75$$
where 
$$\lambda = \frac{\text{torque measured with radial clearance } \delta}{\text{torque at zero clearance}}$$

$$\sigma = \text{relative clearance}$$

$$= \delta/l$$

$$\delta = \text{actual clearance}$$

$$l = \text{height of blade}$$

With the shroud completely removed (rotor without casing, i.e., infinite clearance)  $\lambda \rightarrow 8$ .

Slightly better results can be obtained, if the axes of the nozzles are arranged to lie on a conical instead of a cylindrical surface. Thus an inclination of 10° with the axis of rotation raises  $\lambda_{\sigma=\infty}$  to .84 and  $\lambda_{\sigma=\cdot 05}$  now becomes .955 instead of .940 as for the cylindrical case. It is evident that nozzle inclination reduces the leakage. Since, however, shock losses due to impact of jet with rotor disc are likely to become more pronounced with inclination of jet, the optimum inclination will depend on the particular design investigated and can only be obtained experimentally. In any case the net benefit associated with nozzle inclination is not likely to be large for reasonably small clearances.

Provided that relative clearances of the order  $\sigma=.05$  can be maintained the leakage effect on torque is less than 5 per cent. The author is of the opinion that this slight loss is negligible compared with the advantage due to improved blade cooling made possible by the absence of the shroud and unshrouded blades should therefore always be adopted in gas turbines.

A Photographic Profile Recorder for Airscrews and Model Aerofoil. (R. Kuhl and K. Raab, Luftwissen, Vol. 5, No. 5, May, 1938, pp. 183-185.) (122/3 Germany.)

The aerofoil under examination is supported along its longitudinal axis and subjected to intense transverse illumination by a very narrow plane beam of light. The resulting illuminated section is photographed by a camera which rotates about the specimen, special gearing ensuring that the photographic plate maintains its position in space during the rotation and thus only undergoes a parallel displacement.

The plane of the plate and that of the light used for the illumination are parallel and both are perpendicular to the axis of rotation of the camera and the plate axis. At the same time the optical centre of the objective lies on the line joining the points of intersection "plate plane—plate axis" and "light plane—camera axis of rotation." The optical axis of the camera then lies on a conical surface and since the light beam takes part in the camera rotation, all parts of the contour are photographed in succession and a complete and closed contour is recorded for a camera rotation of 360°. Successive contours along the longitudinal axis of the aerofoil are obtained by displacing the latter longitudinally

through the camera cone, the maximum chord which can be accommodated in the apparatus being limited by the circular path of the objective ( $\sim$  30 cm.). Since the carriage for the longitudinal motion can deal with aerofoils up to 3 m. in length, the apparatus will deal with full-scale propeller blades.

In order to reduce cost, the recorded profile is reduced to half-scale, standard plates  $13 \times 18$  cm. being used in the camera.

Examples of profile distribution along the axis of an airscrew are given. These are photographed on the same plate after successive displacement of the airscrew. By means of reference marks attached to the propeller and reproduced photographically the pitch setting of each section can be determined.

The recorder is also of use for measuring up model aerofoils and checking possible twist or distortion. The profile lines on the plate are very sharp and the co-ordinates can easily be measured to 1/100 mm.

Method of Increasing the Low Temperature Stability of Cellulose Acetate Films. (Z. Rogovin and Z. Ivanova, J. Appl. Chem., U.S.S.R., Vol. 14, No. 6, 1941, pp. 834-842.) (R.T.P. Translation No. 2,121.) (122/4 U.S.S.R.)

The increased brittleness at low temperature  $(-50^{\circ}\text{C.})$  of articles made of cellulose acetate is one of their chief defects and a wide field of application of such products would open out if this difficulty could be overcome.

Whilst it is possible that the low temperature stability or frost resistance of high polymers can be improved by the addition of suitable plasticisers, the authors are of the opinion that a more profitable method of attack is a closer study of the primary material itself and for this reason they have carried out extensive experiments on the effect of molecular size, fractionation, amount of etherification, method of film formation and concentration of acetate on the low temperature stability of unplasticised films, averaging 200  $\mu$  in thickness. The temperature range over which the mechanical properties were tested varied from +80°C. to -50°C., the feature examined being the ultimate tensile strength of the film (kg./mm.²), the percentage elongation at fracture and the number of alternate creasings the film could stand before fracture. The authors consider the latter test as the most valuable in determining the brittleness of the film being much more sensitive than the reduction in percentage elongation. This is borne out by the following table.

Representative figures for a commercial unplasticised film. These were obtained on testing machines specially devised for the purpose.

| Temperature. | Ultimate tensile.       | % Elongation. | Number of repeated creases before fracture. |
|--------------|-------------------------|---------------|---|
| +80°C.       | 7 kg./mm.²              | 20            | 300   |
| + 20°C.      | 9 kg./mm. <sup>2</sup>  | 20            | 150   |
| − 50°C.      | 12 kg./mm. <sup>2</sup> | 10            | 2   |

From the data given by the authors it appears that the principal methods by which low temperature stability of the commercial product can be improved are:—

- (1) Fractionisation, i.e., ensuring that the sample is homogeneous and of high molecular weight ( $\sim$  70,000).
- (2) Reduction in combined acetic acid below 50 per cent.
- (3) High concentration (~ 25 per cent.) of cellulose acetate in solvent (85/15 acetone and alcohol or 85/15 dichlorethane and alcohol).
- (4) Formation of film in vacuo (20-30 mm. Hg., 20°C.).

By combining all the above treatments, it is possible to obtain unplasticised cellulose acetate films the elastic properties of which at  $-50^{\circ}$ C. are at least as good as those of the standard unfractionated film at  $+20^{\circ}$ C.

Vibration Table for Dynamic Tests in the Sonic Frequency Range. (F. L. Meister, A.Z., Vol. 7, No. 2, March, 1942, pp. 51-56.) (122/5 Germany.)

The vibration table developed by the "Reichsanstalt" consists of a small hollow box made of light alloy clamped to the centre of a weak diaphragm. Top and bottom of the box are provided with hollow cylindrical armatures made of high tensile bronze, 10 cm. in diameter and 2 cm. long, and provided with about 10 turns of wire. These armatures project into strong ring-shaped magnetic fields (~ 10,000 gauss) produced by two separately excited electromagnets of suitable shape placed respectively above and below the box. The top armature is supplied with alternating current of accurate sine form and known frequency produced by a valve circuit operating in conjunction with an amplifier and transformer, the bottom armature being used as an indicator of the box motion (induced E.M.F. recorded on an oscillograph).

The apparatus has the following advantages:—

- (1) Very accurate sine vibrations over the frequency range 5-1,0∞/sec. are generated (the lower limit corresponds to the natural frequency of the box diaphragm combination).
- (2) For frequencies below 100/sec., the amplitude can be increased to the order of 1 mm.
- (3) An accurate record of the motion of the table up to the highest frequencies is readily possible.

The fact that the box container is relatively small is no disadvantage, since the modern vibration indicators and accelerometers (for which the test apparatus is mainly intended) are all of very compact design and thus easily accommodated inside the box.

Test results on two carbon pile accelerometers are given. These experiments were carried out at constant vibratory velocity but variable frequency, i.e., the current fed to the forcing coil was so adjusted that the induced E.M.F. on the lower recording coil remained constant and independent of frequency. In this case the accelerometer readings should increase linearly with frequency up to a limit depending on the natural frequency of the instrument and the damping. By repeating the tests with different values for the constant vibratory velocity, the characteristic of the accelerometer as regards natural frequency and damping can be readily obtained. A special point in favour of the vibration table is the excellent cooling of the excitation coil which renders possible endurance runs under stable conditions over considerable periods of time.

Some Notes on the Theory of the Campini System of Jet Propulsion. (G. A. Varshovsky, J. Tech. Physics, U.S.S.R., Vol. 11, No. 12, 1941, pp. 1,123-1,127.) (122/6 U.S.S.R.)

Fundamentally, the Campini system of jet propulsion consists of the following parts:—

- (1) A diverging nozzle, facing in the direction of motion, the entering velocity being  $W_1$ , pressure  $p_1$ , temperature  $T_1$ .
- (2) After compression in this nozzle (at an adiabatic efficiency of  $\eta_d$ ) the air enters an engine-driven compressor at zero velocity, pressure  $p_2$ , where it has its pressure raised to  $p_3$ , the adiabatic compression efficiency being  $\eta_c$ .
- (3) A portion of this air is supplied to the engine, and the remainder, after picking up heat from the engine radiator (or cooling fins) and after further possible heat addition due to the combustion of fuel in an auxiliary combustion chamber, enters the discharge nozzle at a tempera-

ture  $T_4$  and a pressure  $p_3$  (any pressure drop in the flow passages being neglected).

(4) The compressed air finally expands from  $p_3$  to atmospheric pressure  $p_1$ , attaining an exit speed of  $W_2$ .

As is well known, the thrust produced per kg. of air passing through the system  $=(W_2-W_1)/g$  kg., and if the assembly moves in this direction through still air, the propulsive work per kg.  $=W_1\,(W_2-W_1)/g$  kgm. Dividing this by q (the total amount of heat added per kg. of air) gives the thermal efficiency  $\eta_e$  of propulsion whilst division by the kinetic energy at entry gives what the author calls the thrust coefficient C=2  $(W_2-W_1)/W_1$ .

The author obtains expressions for  $\eta_e$  and C which are functions of the following quantities:—

- (1) The Mach number at entry.
- (2)  $\gamma$  and  $C_{\rm p}$  of air (it is assumed that the products of combustion have the same specific heat).
- (3)  $\eta_d$  and  $\eta_c$ .
- (4)  $\eta_m$ =thermal efficiency of engine.
- (5)  $\phi$  = adiabatic efficiency of reaction nozzle.
- (6)  $\psi$ =percentage heat utilisation factor ( $i-\psi$ =percentage heat loss of system).
- (7) 1/n = percentage of total heat supplied to engine (r 1/n = heat supplied to auxiliary combustion chamber).
- (8)  $T_1 = \text{temperature at entry.}$
- (9) Total heat supplied q.

Keeping the factors (2) to (7) constant,  $\eta_e$  is thus a function of Ma, q and  $T_1$  only and curves are plotted for  $\eta_e$  as a function of  $q/C_pT_1$  with  $Ma^2$  as parameter, using the following constants:—

$$\eta_0 = .96$$
 $\eta_0 = .88$ 
 $\phi = .98$ 
 $\psi = .98$ 
 $\eta_m = .29$  and .40 respectively
 $\eta_n = 1$ 

With  $\eta_{\rm m}$ =.29 and a propeller efficiency of 80 per cent., a standard engine will give a thermal propulsion efficiency of about 23 per cent. The curves show that the Campini system will exceed this at  $Ma^2$ =.2 for  $q/C_{\rm p}T_1$  between .1 and .2 and for  $Ma^2$ =.4,  $\eta_{\rm e}$  will be in excess of 25 per cent. over the  $q/C_{\rm p}T_1$  range .1 to .5.

For higher Ma values, the benefits of the jet system become more pronounced. When applied, however, to an engine of higher inherent thermal efficiency  $(e.g., \eta_n = .4)$  Mach numbers of .6 will have to be exceeded before the jet system discussed shows any marked improvement over the standard propeller. (N.B.—It is assumed that the propeller can be designed to maintain 80 per cent. efficiency throughout.)

Experimental Layout for Recording Speed Variation in Turbulent Flow. (G. Datwyler, Mitt. E.T.H., Zurich, No. 8, 1943, pp. 34-43.) (122/7 Switzerland.)

Airspeed measurements with a hot wire anemometer consist in recording the voltage drop across the wire for a constant heating current. On account of thermal inertia, such a wire does not respond immediately to speed fluctuations

and the record is distorted both as regards amplitude and phase. If  $a_{\rm w} = {\rm response}$  to a speed amplitude a at frequency  $w = 2\pi f$ .  $a_{\rm o} = {\rm response}$  under steady conditions  $w = {\rm o}$ ,

we have

$$\frac{a_{\mathbf{w}}}{a_{\mathbf{o}}} = \frac{1}{\sqrt{(1+w^2M)}} e^{-i \tan^{-1}wM} \qquad . \tag{1}$$

where M = time constant of an emometer.

$$= \frac{mS \ (T-T_{\rm o})}{(.24 \ I^2 R_{\rm o})}$$

where m = mass of wire.

S = its specific heat.

T = mean temperature of wire.

T<sub>o</sub> = temperature of unheated wire.

 $R_{o}$  = resistance of unheated wire.

• I = current.

M thus represents the time required for the wire to reach the working temperature T with a constant heat input of .24  $I^2R_o$ , starting at  $T_o$ .

Equation (1) shows that under these conditions, the recorded amplitude is decreased in the ratio  $1/\sqrt{(1+w^2M^2)}$  and there is a phase lag of  $\tan^{-1}wM$  in addition.

If the instrument is therefore to be used for turbulence investigations, the electric recording circuit must be such that the amplification increases with frequency and that the phase is automatically advanced.

This is achieved in a relatively simple manner by putting into the plate circuit a resistance R in series with an inductance L. Under these conditions, the amplification  $\nu$  becomes

$$v = \{ \frac{\mu}{(\mathbf{I} + R_i/R)} \}$$

$$\sqrt{\left[ \{ \mathbf{I} + (wL/R)^2 \} / \{ \mathbf{I} + (wL)/(R_i + R)^2 \} e^{i \tan^{-1}\left[(wL/R)\{1 - 1/(1 + Ri/R)\}\right]} \right]}$$
where  $R_i$ =internal resistance of valve
$$\mu = \text{no load amplification of valve}.$$

This is the correct amount, provided

$$L/R = M$$
.  
 $R_i \gg R$ .  
 $\mu = \text{constant over frequency range.}$ 

In practice,  $R_i$ , although large, cannot be considered as  $\infty$ . The resulting error is however very small. Thus in a particular case

$$R_1 = 750,000 \ \Omega.$$
 $L = 3$  Henry.
 $M = .004$  sec.
 $R = 750 \ \Omega.$ 
 $f = 10$  100 1,000 10,000
 $1/\sqrt{\{1 + (wL/R_1 + R)^2\}} = 1$  1 1 1.031

The amplitudes are thus reproduced up to frequencies of 10,000/sec. with an error of less than 3 per cent., whilst the phase error over the same range is negligible.

The arrangement adopted in the E.T.H. Laboratory consists of a three-stage amplifier with capacity coupling, the first and last stage acting as normal resistance amplifiers whilst the intermediate stage is provided with the inductive compensation described above, together with a five-step grid potentiometer in order to adjust the amplification to the degree of existing turbulence.

The anode circuit of the last stage is provided with a sensitive thermo ammeter, consisting of a thermocouple attached to a fine resistance. The E.M.F. generated in the couple varies as the square of the anode current and thus is a measure of the mean square speed fluctuation of the flow (independent of frequency). The actual speed fluctuations can also be recorded on an oscillograph.

As an example of the application of the apparatus, turbulence surveys in the wake of a model aerofoil of 340 mm. chord are given for various distances from the trailing edge. Defining the turbulence factor as  $\sqrt{u^2/U}$  when U = undisturbed stream flow, the records show a maximum value of 7.8 per cent. at 30 mm. and dropping to 5 per cent. at 100 mm.

Striation Technique Applied to the Supersonic Wind Tunnel. (P. de Haller, Mitt. E.T.H., Zurich, No. 8, 1943, pp. 44-49.) (122/8 Switzerland.)

The great advantage of striation technique when applied to high speed flow phenomena is the fact that the field of flow can be surveyed qualitatively without creating additional disturbances due to the introduction of measuring instruments into the high speed air stream. The application of the method to supersonic wind tunnels is, however, complicated by the large field of view required (spherical aberration of optical system!) as well as by the deformation of the observation windows due to the relatively high vacuum usually existing in the tunnel. To overcome these difficulties, the whole system comprised of light source, slot, edge plate and camera is mounted on a common stand and can be orientated in any direction.

The wind tunnel measuring section is provided with two windows and carries a concave mirror of 3 m. focal length facing the rear window. The light (Phillips high pressure mercury vapour lamp) after being focussed on the slot traverses the channel section from front to rear (40 cm.) and on emergence at the back window (30 cm. diameter) is reflected by the concave mirror, thus producing an image of the slot on the edge plate. The light then passes through the camera objective and is reflected by a plane mirror on to a ground glass screen or photographic plate placed close to and slightly to one side of the camera lens. The position of the edge plate relatively to the illuminated slot can be adjusted by means of a micrometer screw, whilst their common mounting can be rotated so as to observe density gradients in different directions. The lens system, being highly achromatic, enables the use of a powerful light source without the need of a filter. The fact that the light traverses the measuring field twice increases the sensitivity of the method still further.

Although, as already stated, the striation method is intended mainly for qualitative studies of the field of flow, it is interesting to note that the apparatus enables quantitative measurements of pressure gradients of the order of 1 per cent. per cm. For this purpose, a number of subsequent exposures are taken for different positions of the edge plate, the exposure times and air pressure being the same as in the flow investigation proper, with the exception that the wind velocity is now zero. These exposures are made on a previously unexposed portion of the film. After development, the density of these exposures is compared with those of the flow pattern by means of a photometer. This enables the corresponding angles of deviation to be determined irrespective of difference in emulsion or development of film.

The Aerodynamic Laboratory of the E.T.H., Zurich. (J. Ackeret, Mitt. E.T.H., No. 8, 1943, pp. 5-33.) (122/9 Switzerland.)

The E.T.H. wind tunnel has a working section of  $3 \times 2.1$  m. which is adjustable, so that the tunnel walls can be made either convergent, divergent or parallel.

In this manner a constant pressure can be maintained over an appreciable length of tunnel which is very important in drag investigations on fuselage models. On the other hand, experiments on turbine blades can be carried out with a negative pressure gradient and actual working conditions thus reproduced more accurately. Finally the tunnel walls can be removed entirely and the wind tunnel operated with a free jet, but adjustable collector nozzle. This is of advantage when relatively large models have to be investigated, since in this case the velocity distribution is less interfered with (displacement flow). The adjustment of the collector nozzle proved very beneficial in reducing stagnation effects and renders the subsequent velocity conversion very efficient, the overall tunnel factor being 3.86 (referred to power supply).

The air current is supplied by two axial blowers (2.5 m. blade diameter) situated in the return circuit and operating in parallel. Each of these blowers is rated at 275 h.p. at 1,400 r.p.m. and supplies 230 m.³/sec. at a pressure difference of 80 kg./m.². At this continuous rating, the air speed in the working section is of the order of 80 m./sec., which can be increased to 90 m./sec. for short periods (20 minutes). The D.C. motors driving the fans are under Ward-Leonard control, speed synchronisation being assured by a silk belt. Considerable trouble was at first experienced with variations in the supply frequency of the three-phase motor driving the Leonard group. The corresponding speed variations of the driving motor was finally cured by an electric valve circuit which controls the excitation of the dynamo to within very narrow limits.

In addition to this large tunnel, the laboratory also has a small high speed tunnel with a circular working section of 25 cm. diameter. This apparatus is mainly intended for the calibration of pitot tubes at speeds up to 170 m./sec.

The most interesting equipment of the E.T.H. is supersonic wind tunnel with a working section of  $40 \times 40$  cm. By operating in a closed circuit under low pressure, a Mach number of 2 can be reached for a power consumption of less than 1,000 h.p. The tunnel is operated by a 13-stage axial compressor supplied by B.B.C. and running at a maximum speed of 3,900 r.p.m. This type of machine was chosen, since its restricted diameter (~ 120 cm.) facilitates installation. Moreover, control is facilitated, since by cutting out some of the guide wheels, the pressure ratio can be reduced from its maximum value of 2.4 without the idle stages absorbing appreciable power.

Elaborate precautions are taken to pressure seal the bearings of the compressor and prevent oil entering the air stream.

As is well known, cooling the air of a supersonic tunnel presents a major problem. A special air cooler capable of dealing with 750,000 K. cal./hour is built into the circuit preceding the working section. The cooler consists of three independent units placed one behind the other, and although offering an appreciable resistance, its presence serves to steady down the air stream.

Apart from the usual investigations on aerofoils, the tunnel has lately been used for experiments on a 300 h.p. air turbine. It appears that a reduction in the aerodynamic losses of such machines ranks at least equal with metallurgical improvements. By inserting the turbine in the supersonic wind tunnel circuit, the effect of variation of Re and Ma on performance can be readily studied. Moreover, by means of special apparatus, it is hoped to follow out the flow in the interior of the machine. According to the director of the E.T.H. such turbines are likely to become the power plant of the future and such a development by itself would more than justify the expense of the supersonic tunnel.

Development of Stall Warning Indicators. (J. George, 12th Annual Meeting, Inst. Aeron, Sciences, Jan., 1944.) (Preprint available.) (122/10 U.S.A.)

From statistics it appears that over 100 fatal accidents directly attributable to stalling occurred in the U.S.A. in 1941. There is thus an urgent need for a

reliable and practical device which will warn the pilot in a very positive manner of the approaching stall. Obviously, utmost reliability is essential, since, once adopted, pilots would depend on the warning and a faulty response would constitute an increased hazard. Experiment has shown that for a given aerofoil, stalling occurs at a definite angle of attack irrespective of air speed, and the problem is therefore solved if a reliable angle of attack indicator can be evolved. Two successful devices of this type have been investigated by the C.A.A. The first. known as the Gurley indicator, depends on the travel of the stagnation point at the leading edge of the wing as the incidence changes. Under normal flight conditions this point is on the upper surface of the wing, but will move round the nose on to the lower surface with increasing angle of attack (stall). light vane projecting from the nose into the air stream will be deflected downwards if the stagnation point is above the vane. This corresponds to normal flight conditions. At minimum safe flying speed, the stagnation point has travelled below the vane and will deflect it upwards, thus closing a switch operating a klaxon or signal lamp. In practice, the position of the vane is chosen so that a warning is given at a speed of about 25 per cent. in excess of stalling speed. Templates are provided so that this position can be readily found on any given shape wing. It is stated that flight tests have proved satisfactory and the instrument, which is simple and robust, is recommended for the private owner. Only one instrument, situated midway between fuselage and wing tip need be fitted.

The second angle of attack indicator tested by the C.A.A. (Wayne University type) depends on the change in pressure distribution on the nose with change in incidence. At any angle of attack there are two points on the leading edge contour, which are at the same pressure as the interior of the wing. increase in incidence, these points of zero pressure difference travel downwards around the nose in such a manner that the upper portion of the leading edge suffers a pressure reversal from positive to negative, whilst the reverse holds for the lower portion. A diaphragm connected to a point on the upper surface will thus undergo a change in deflection at the angle of attack corresponding to minimum safe flying speed. By utilising a slack diaphragm of rubber-like material sufficient deflection to operate an electrical contact can be obtained with a pressure difference of the order of .015 in. of water. Although slightly more complicated than the Gurley indicator, the instrument is less subject to damage since there are no projecting parts. The sensitivity is slightly greater than that of the Gurley indicator. Both the above angle of attack indicators will obviously only function in the absence of icing, since otherwise the wing contour and hence the calibration changes. Stall warning devices operating under conditions of icing are more difficult to achieve since they will have to respond to changes in the character of the flow over the whole wing surface and will necessitate simultaneous readings of pressure differentials at three or more points situated near the trailing edge. Although promising laboratory experiments have been carried out, this type of indicator still awaits flight tests under actual icing The fundamental requirement of any stall indicator (pressure differential independent of air speed and function of angle of attack only) appears to be very difficult to achieve under icing conditions, and this may delay the advent of a suitable instrument although urgently needed by civil aviation transport.

Device for Stripping Insulation from Electric Conductors (D.R.G.M. 1,504,595). (Der Flieger, Vol. 22, No. 7, July, 1943, pp. 211-212.) (122/11 Germany.)

The stripping of insulation by hand, using a knife, pliers, or similar tools is unsatisfactory, since it takes an appreciable time and fatigues the operator. The alternative of burning off the insulation by means of a red hot wire has also not

proved amenable to mass production conditions. For this reason the Junkers firm have developed a mechanical stripping machine, consisting of a clamping device (upper and lower straight edge) provided with a series of recesses of standard diameter corresponding to the types of cable requiring trimming. Immediately behind the clamp is situated a cutter, consisting of an upper and lower knife which surrounds the cable. The travel of the cutter is controlled by the thickness of the insulation, and after cutting through to the metal strands, the knife travels back along the cable, stripping off the insulation automatically.

#### LIST OF SELECTED TRANSLATIONS.

#### No. 68.

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Lists of selected translations have appeared in this publication since September, 1938.

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|       | •                        |        | _     |       |       |         |
|-------|--------------------------|--------|-------|-------|-------|---------|
|       | Index.                   |        |       |       |       | Items.  |
| I.    | Theory and Practice o    | f War  | fare  | • • • |       | 1-192   |
| H.    | Aerodynamics and Hyd     | rodyna | mics  | •••   |       | 193-206 |
| III.  | Aircraft and Accessorie  | s      |       |       |       | 207-324 |
| IV.   | Engines and Accessorie   | s      | • • • |       |       | 325-399 |
| v.    | Fuels and Lubricants     | •••    | • • • |       |       | 400-439 |
| . VI. | Theory of Elasticity     | • • •  | • • • | • • • |       | 440-460 |
| VII.  | Materials                | • • •  |       |       |       | 461-682 |
| VIII. | Instruments              | • • •  | •••   |       |       | 683-707 |
| IX.   | Production               | • • •  | •••   | • • • |       | 708-844 |
| X.    | Transport                | • • •  |       |       |       | 845-868 |
| XI.   | Wireless and Electricity |        | •••   |       |       | 869-904 |
| XII.  | Sound, Light and Heat    | •••    |       |       |       | 905-921 |
| XIII. | Photography              |        | •••   |       |       | 922-927 |
| X·IV. | Meteorology              | • • •  |       |       | •••   | 928-933 |
| XV.   | Physiology and Aviation  | n Medi | cine  | •••   | • • • | 934-975 |
| XVI.  | Mathematics              | •      | •••   | ···   |       | 976-982 |
|       |                          |        |       |       |       |         |

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|------|-------|--------|---|--|
| NO.  | H     | EF.    |   | TITLE AND JOURNAL.   |
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              U.S.A.
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120
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      19929
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                       Military Types of Aircraft (Germany).
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              Germany
                               (Flight, Vol. 45, No. 1,829, 13/1/44, pp. 42-43.)
Bucker Primary Trainer Bu 182 "Kornett."
     18643
129
              Germany
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Gotha GO. 146 Twin-Engined Light Transport.
130
     18644
              Germany
                                  (Flugsport, Vol. 35, No. 16, 15/12/43, pp.
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             Germany
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     18737
              Germany
134
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              Germany
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                                  (Photo). (Flight, Vol. 45, No. 1,832, 3/2/44, p.
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| 198        | 19072 | G.B          |     | 1943, pp. 1-16.) Flow States in Emergent Gas Streams. (J. E. Garside and others, Nature, Vol. 152, No. 3,869, 25/12/43, p. 748.)   |
| 199        | 19370 | G.B          | ••• | Possibilities in Boundary Layer Control. (Flight, Vol. 45, No. 1,832, 3/2/44, p. 130.)   |
| 200        | 19680 | Germany      |     | The Theory of the Steady Compression Shock. (Z.V.D.I., Vol. 87, No. 51-52, 25/12/43, pp. 818-819.)   |
| 202        | 19710 | U.S.A.       |     | Aeronautical Index for 1939 (a Subject and Author<br>Index to Aeronautical Periodicals and Technical<br>Reports). (Institute of Aeronautical Sciences,<br>1943, pp. 1-315.)                                  |
| 203        | 19727 | G.B          |     | The Influence of Reynolds Number at High Mach<br>Numbers. (A. Ferri, J. Roy. Aeronaut. Soc.,<br>Vol. 48, No. 398, Feb., 1944, pp. 45-48.)  |
| 204        | 19869 | U.S.A.       |     |  |
| 205        | 19882 | Germany      |     | Influence of Re on M as Shown by Experiments on Spheres and Cylinders. (Digest of Atti di Guidonia 67-69.) (Profile series No. 39.) (H. O. Nicolaus, Flugsport, Vol. 36, No. 1, 19/1/44,                     |
| 206        | 19920 | G.B          | ••• | pp. 157-160.) Shock Waves at Sonic Speeds (Abstract). (W. F. Hilton, Flight, Vol. 45, No. 1,834, 17/2/44, p. 182.)   |

## AIRCRAFT, AIRSCREWS AND ACCESSORIES. Airlines and Civil Air Transport.

207 18525 U.S.A. ... Airways for Peace (Survey of International Air Transport Problem). (Edward Warner, Foreign Affairs, Vol. 22, No. 1, Oct., 1943, pp. 11-27.)

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| 208         |       | U.S.A.         |        | Post-War Private Flying. (J. H. Geisse, S.A.E.  |
|             |       |                |        | Preprints, 10-14/1/44, pp. 1-20.)   |
| 209         | 18569 | G.B            | •••    | Post-War Transport in Great Britain. (B. D. Richards, Mechanical World, Vol. 114, No. 2,969,  |
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|             |       | C D            |        | 100-104.)   |
| 214         | 19200 | G.B            |        | Indian Civil Aviation. (Aeronautics, Vol. 9, No. 5, Dec., 1943, p. 52.)   |
| 215         | 19289 | G.B            |        | Post-War Transport in Great Britain. (D. B. Richards, Civil Engineering, Vol. 38, No. 450, Jan., 1944, pp. 266-267.)  |
| 216         | 19293 | G.B            |        | Flying Boats in Post-War Aviation. (O. Stewart, Aeronautics, Vol. 9, No. 6, January, 1944, pp.  |
| 217         | 19497 | G.B            | •••    | 27, 51-53.) China's Life-Line (the Air Route from Bengal to China). (Aeroplane, Vol. 66, No. 1,706, 4/2/44,   |
| 218         | 19503 | U.S.A.         |        | p. 132.) Britain's Pioneer Work in the Development of African Air Routes. (Aviation, Vol. 42, No. 11, November, 1944, pp. 196-197, 281-290.)  |
| 219         | 19775 | G.B            |        | The Future of Air Transport in Europe—I. (General M. Valin, Aeroplane, Vol. 66, No. 1,707, 11/2/44,   |
| 220         | 19836 | U.S.A.         |        | pp. 165-166.) The Philosophy of Aviation. (R. S. Damon, Mechanical Engineering, Vol. 66, No. 1, Jan., 1944, pp. 13-14, 71.)   |
|             |       | Civ            | il and | l Experimental Aircraft Types.  |
| 221         | 18512 | U.S.A.         |        | Nash-Kelvinator Sikorsky Helicopter (Photo).  |
| ~~1         | Ü     | -              | 777    | (Iron Age, Vol. 151, No. 24, 17/6/43, p. 78.)<br>Horten Va. Tailless Aircraft Utilising Phenolic  |
| 222         | 18646 | Germany        |        | Horten Va. Tailless Aircraft Utilising Phenolic<br>Resins as Structural Material (Engine Bearers<br>and Undercarriages Steel Tubing). (Horten,<br>Flugsport, Vol. 35, No. 16, 15/12/43, pp. |
|             | 06    | C              |        | 249-251.)   |
| 223         | 18050 | Germany        |        | Horten IV Tailless Glider (Photograph Showing<br>Prone Position of Pilot). (Flugsport, Vol. 35,<br>No. 16, 15/12/43, p. 259.)   |
| 224         | 19033 | Canada         |        | V.S. 300 Helicopter. (Canadian Aviation, Vol. 16, No. 5, May, 1943, pp. 62-66, 120.)  |
| 225         | 19057 | Canada         |        | Powered Gliders—Sky Freighters of the Future. (K. Edger, Canadian Aviation, Vol. 16, No. 9,   |
| 226         | 19174 | G.B            | •••    | Sept., 1943, pp. 52-53.)  A Successful Rotating Wing Aircraft (Sikorsky).  (Aeronautics, Vol. 9, No. 3, October, 1943, p. 57.)  |

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| 228         | 19379 | G.B            | •••     | The Avro York. (Aeroplane, Vol. 66, No. 1,705, 28/1/44, pp. 105-106.)   |
| 229         | 19510 | U.S.A.         | •••     | PV-2 Helicopter Makes First Public Flight. (Aviation, Vol. 42, No. 11, November, 1943, p. 229.)   |
| 230         | 19750 | G.B            | •••     | Flying Boats. (A. Gouge, Flight, Vol. 45, No. 1,833, 10/2/44, pp. 152-153.)   |
| 231         | 19776 | France         | •••     | Bloch 161 and 220 Air Liners (Photos). (Aeroplane, Vol. 66, No. 1,707, 11/2/44, p. 146.)  |
| 232         | 19879 | Germany        | •••     | British and American Cargo Gliders. (Flugsport, Vol. 36, No. 1, 19/1/44, pp. 2-4.)  |
| 233         | 19897 | Germany        | •••     | Blohm and Voss B.V. 138 Long Range Flying Boat. (Flugsport, Vol. 35, No. 15, 17/11/43, pp. 225-226.)  |
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| 234         | 18548 | G.B            | •••     | Weight Estimating. (L. W. Rosenthal, Flight, Vol. 45, No. 1,829, 13/1/44, p. 41.)   |
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| 238         | 18902 | U.S.A.         |         | Use of Wood in Aircraft Design and Construction—Part I. (J. J. Wallace, Aero Digest, Vol. 43, No. 5, November, 1943, pp. 190-191, 284.)   |
| 239         | 18913 | U.S.A.         | •••     | Weight Control in Specification Writing—Part 1.<br>(J. E. Ayers, Aero Digest, Vol. 43, No. 5.<br>November, 1943, pp. 251-252.)  |
| 240         | 19051 | Canada         |         | Standard Sizes of Aircraft Tubing. (Canadian Aviation, Vol. 16, No. 8, Aug., 1943, p. 140.)   |
| 241         | 19159 | G.B            | ···     | Plane Vans (Detachable Fuselage). (Aeronautics, Vol. 9, No. 4, November, 1943, p. 68.)  |
| 242         | 19194 | G.B            |         | Structural Integration (Pertinent Criticisms of<br>Modern Aircraft Design). (Aeronautics, Vol. 9,<br>No. 5, Dec., 1943, p. 27.)   |
| 243         | 19255 | U.S.A.         | • • • • | Aircraft Electric Remote Controls. (W. P. Lear, S.A.E. Preprints, 16/4/43.)   |
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| 246              | 19305 | G.B         | •••   | Standardised Flight Controls (General Aircraft, Ltd., Patent). (Aeronautics, Vol. 9, No. 6, Jan., 1944, pp. 62-63.)  |
| 247              | 19388 | G.B         | •••   | Colour and Design in Civil Aviation. (Flight, Vol. 45, No. 1,831, 27/1/44, p. 104.)  |
| 248              | 19471 | U.S.A.      | •••   | Deficiencies of Converted Passenger Aeroplanes for Cargo Transport and Operating Requirements. (C. Froesch, S.A.E. Journal, Vol. 51, No. 12, December, 1943, pp. 432-438.)   |
| 249 <sup>.</sup> | 19477 | U.S.A.      | •••   | Some Tactical Considerations in Aircraft Design (Summary of Paper). (F. O. Carroll, S.A.E. Journal, Vol. 51, No. 12, December, 1943, pp. 33, 35-36.)   |
| 250              | 19916 | G.B         |       | Tail of Campini Jet Propelled Monoplane is Completely Detachable (Photo). (Flight, Vol. 45, No. 1,834, 17/2/44, p. 172.)   |
|                  |       | Perfo       | orma  | nce (Calculations and Testing).  |
| 251              | 18540 | U.S.A.      | •••   | Structural Flight Research (Determination of Static and Manœuvring Loads, etc.). (W. L. Howland, S.A.E. Preprints, 10-14/1/44, pp. 1-10.)  |
| 252              | 18557 | Germany     | •••   | Comparison of Various Methods for Determining<br>Flight Speed at Great Altitudes. (R.T.P.3 Trans-<br>lation No. 1,797.) (R. Schmidt, Journal of the<br>Royal Aeronautical Society, Vol. 48, No. 397,<br>January, 1944, pp. 12-23.)       |
| 253              | 18673 | Germany     | •••   | A New Low Temperature Test Chamber (Temperatures Down to -115°C.) (Digest). (From Zeitschrift für die gesamte Kälte-Industrie, Vol. 50, Nos. 5-6, May-June, 1943, pp. 57-62.) (H. Glaser, Engineers' Digest, Vol. 4, No. 12, Dec., 1943, |
| <sup>2</sup> 54  | 18698 | Switzerland | •••   | PP. 343-347.) Some Simplified Methods for Determining Diving Speeds (Digest). (J. Patry, Flugwehr und Technik, Vol. 5, No. 3, March, 1943, p. 78.)   |
| 255              | 18897 | U.S.A.      |       | Performance Calculations (Study of Drag). (M. M. Munk, Aero Digest, Vol. 43, No. 5, November,  |
| 256              | 19018 | U.S.A.      |       | 4943, pp. 160-162, 304-320.)  Record-Breaking Non-Stop Flight by Gliders. (U.S. Air Services, Vol. 28, No. 9, Sept., 1943, p. 38.)   |
| 257              | 19036 | Canada      | •••   | Atlantic Speed Records Set Up by Ferry Command<br>Planes. (Canadian Aviation, Vol. 16, No. 5,<br>May, 1943, p. 88.)  |
| 258              | 19185 | G.B         | •••   | Initial Performance Testing. (R. G. Worcester, Aeronautics, Vol. 9, Nos. 4-6, August, 1943,  |
| 259              | 19883 | Germany     | •••   | pp. 44-46.) Competition for Engine Driven Model Aircraft (List of Regulations). (Flugsport, Vol. 36, No. 1, 19/1/44, pp. 15-16.)   |
| 260              | 19917 | G.B         | •••   | Flight Testing Methods (Abstract of Paper to R. Aer. Society). (E. T. Jones, Flight, Vol. 45, No. 1,834, 17/2/44, pp. 175-176.)  |

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|      |        |               | Stability, Control.   |
| 261  | 18690  | Switzerland.  | The Take-off of Heavily Loaded Aircraft. (H. 1. Studer and F. Widmer, Flugwehr und Technik, Vol. 5, No. 3, March, 1943, pp. 75-77.)   |
| 262  | 18694  | Switzerland . | The Take-off of Heavily Loaded Aircraft. (H. L. Studer and F. Widmer, Flugwehr und Technik, Vol. 5, No. 3, March, 1943, pp. 75-77.)   |
| 263  | 18708  | Switzerland.  | Wing Forces Due to Torsion (Deformation of Section). (A. Manser, Flugwehr und Technik, Vol. 5, No. 1, Jan., 1943, pp. 21-23.)   |
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| 265  | 19146  | G.B           | The Flutter Problem (Some Early Attempts at Prevention). (Aeronautics, Vol. 9, No. 4, November, 1943, p. 27.)   |
| 266  | 19860  | Canada .      | 77' 7 C 7 C/ 77 /7 C1 C 1' A '-'  |
|      |        |               | General Equipment.  |
| 267  | 18844  | U.S.A         | Supplementary Fuel Tanks (Made of Phenolic Sisal). (Modern Plastics, Vol. 21, No. 2, October, 1943, p. 78.)   |
| 268  |        | U.S.A         | Lightweight Hinge for Aircraft (Plastic). (Modern Plastics, Vol. 21, No. 2, October, 1943, p. 79.)  |
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| 270  | 18854  | U.S.A.        | Captured Enemy Material (Use of Plastics in A.A. Equipment, Guns, etc.). (Modern Plastics, Vol. 21, No. 2, Oct., 1943, pp. 80, 130.)  |
| 271  | 18898  | Ú.S.A         | Parallel Operation of Aeroplane Alternators. (D. W. Exner, Aero Digest, Vol. 43, No. 5, November, 1943, pp. 172-174, 298-304.)  |
| 272  | 19324  | G.B           | Anti-Vibration Foundation for Air Compressor. (Machinery, Vol. 63, No. 1,627, 16/12/43, pp. 678-679.)   |
| 273  | 19609  | U.S.A         | New Aircraft Heater for High Altitude Flying. (American Aviation, Vol. 7, No. 14, 15/12/43, p. 72.)   |
|      |        |               | Propellers.   |
| 274  | 10306  | G.B           |   |
| ~/-  | 19300  |               | Airscrews, Ltd., Patent). (Aeronautics, Vol. 9, No. 6, Jan., 1944, p. 63.)  |
| 275  | 19667  | Germany .     |   |
| 276  | 19884  | Germany .     | Vol. 87, Nos. 5-6, 6/2/43, p. 86.)  Compressed Wooden Airscrew Blades (728,750).  (Heine, Flugsport, Vol. 36, No. 1, 19/1/44, p. 81.)   |
| 277  | 19885  | Germany .     | Constant Speed Varrell Pitch Airscrew Mechanism, including Brake Operation (738,969). (Junkers, Flugsport, Vol. 36, No. 1, 19/1/44, p. 81.)   |

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| 278         |       | Germany       |       | TITLE AND JOURNAL.  Epicyclic Gear for Airscrew Blades (739,833).  (V.D.M., Flugsport, Vol. 36, No. 1, 19/1/44, p. 81.)                              |
| 279         | 19887 | Germany       | •••   | Hydraulic V.P. Airscrew Mechanism (739,218). (Junkers, Flugsport, Vol. 36, No. 1, 19/1/44, p. 82.)   |
| 280         | 19888 | Germany       |       | Blade Stops for Hydraulically Operated V.P. Airscrew (739,834). (V.D.M., Flugsport, Vol. 36, No. 1, 19/1/44, p. 82.)                                 |
| 281         | 19889 | Germany       | •••   | Cooling Air Intake Control for Air-Cooled Engine<br>Operating V.P. Airscrew (739,719). (B.M.W.,<br>Flugsport, Vol. 36, No. 1, 19/1/44, pp. 82-83.)   |
| 282         | 19894 | Germany       |       | Utilising Propeller Inertia for Starting Up Aero<br>Engine (738,877). (D.B., Flugsport, Vol. 36,<br>No. 1, 19/1/44, pp. 84-85.)                      |
| 283         | 19907 | Germany       |       | Automatic Reverse Pitch Propeller Braking Operated by Ground Contact Pressure (736,112). (Heinkel, Flugsport, Vol. 35, No. 15, 17/11/43, pp. 70-71.) |
| 284         | 20057 | Switzerland   | •••   | Note on "Use of Propeller as a Brake." (J. Ackeret, Flugwehr und Technik, Vol. 5, No. 3, March, 1943, p. 83.)  |
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| 285         | 18849 | U.S.A.        | •••   | Transparent Enclosures in Aircraft. (O. E. Brown and W. N. Arata, Modern Plastics, Vol. 21, No. 2, Oct., 1943, pp. 87-93, 140.)                      |
| 286         | 19176 | G.B           | • • • | Pressure Cabin (Bendix Patent). (Aeronautics, Vol. 9, No. 3, October, 1943, pp. 58-59.)  |
| 287         | 19203 | G.B           | •••   | A Gallery of Cockpits (Photos). (Aeronautics, Vol. 9, No. 5, Dec., 1943, pp. 58-59.)   |
|             |       |               |       | Wings, Flaps, etc.   |
| <b>28</b> 8 | 18651 | Germany       | •••   | Retractable and Rotatable High Lift Flap (738,636). (Arado, Flugsport, Vol. 35, No. 16, 15/12/43, p. 75.)  |
| 289         | 18665 | Germany       | •••   | Aerodynamic Wing Brake (715,740). (Messerschmitt, Flugsport, Vol. 35, No. 16, 15/12/43, p. 80.)  |
| 290         | 19204 | G.B           |       | The Youngman Wing Flap (Fairey Patent). (Aeronautics, Vol. 9, No. 5, Dec., 1943, p. 60.)   |
| 291         | 19908 | Germany       |       | Combined Diving Brake and Landing Flap (737,178). (D.F.S., Flugsport, Vol. 35, No. 15, 17/11/43, p. 71.)   |
| 292         | 19909 | Germany       | •••   | Diving Brake Consisting of Staggered Prongs (737,445). (Arado, Flugsport, Vol. 35, No. 15, 17/11/43, p. 71.)   |
|             |       |               |       | Landing Gear.  |
| 293         | 18552 | G.B,.         |       | Tricycle Landing Gears—Review of Design and<br>Performance Criteria. (Flight, Vol. 45, No.<br>1,829, 13/1/44, pp. 48-49.)                            |
| 294         | 18660 | Germany       | •••   | Spring Legs and Shock Absorbers (739,677-739,678). (Elektron, Flugsport, Vol. 35, No. 16, 15/12/43, p. 76.)  |

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| NO.  | :        | REF.    |           | TITLE AND JOURNAL.   |
| 295  | 18661    | Germany |           | Spring Legs (740,079). (Messerschmitt, Flugsport, Vol. 35, No. 16, 15/12/43, p. 78.)   |
| 296  | 18662    | Germany |           | Retractable Undercarriage (739,679). (Henschel, Flugsport, Vol. 35, No. 16, 15/12/43, pp. 78-79.)  |
| 297  | 18663    | Germany | •         | Retractable Undercarriage (739,680). (Messerschmitt, Flugsport, Vol. 35, No. 16, 15/12/43, p. 79.)   |
| 298  | 18664    | Germany |           | Jettisonable Undercarriages (Flexible Wheel Connection) (740,220). (Gotha, Flugsport, Vol. 35, No. 16, 15/12/43, p. 80.)                             |
| 299  | 18826    | G.B     | •••       | Some Notes on Pneumatic Tyres. (R. Hadekel, Aircraft Engineering, Vol. 16, No. 179, January,   |
| 300  | 19901    | Germany | • • • •   | 1944, pp. 11-13, 17:)  Preliminary Rotation of Landing Wheels (735,458).  (Elektron, Flugsport, Vol. 35, No. 15, 17/11/43, p. 65.)                   |
| 301  | 19902    | Germany |           | Undercarriage Dampers (735,460, 736,718). (Elektron, Flugsport, Vol. 35, No. 15, 17/11/43, pp. 65-66.)   |
| 302  | 19903    | Germany | • • •     | Ring Springs for Undercarriage Struts (737,615). (Focke-Wulf, Flugsport, Vol. 35, No. 15, 17/11/43, p. 66.)  |
| 303  | 19905    | Germany | •         | Retractable Undercarriages (735,188, 735,942, 736,110, 736,720). (Various, Flugsport, Vol. 35, No. 15, 17/11/43, pp. 67-69.)                         |
| 304  | 19906    | Germany | •••       | Landing Gear for Either Skid or Wheel Operation at Will (736,719). (Arado, Flugsport, Vol. 35, No. 15, 17/11/43, p. 68.)                             |
|      |          |         |           | De-icing.  |
| 305  | 18922    | U.S.A.  | •••       | De-Icer System with Electronic Control. (Aero Digest, Vol. 43, No. 5, November, 1943, p. 324.)   |
| 306  | 19031    | Canada  | •••       | Lumarith Plastic Sheets for Windshields (Solves Icing Problem on Seaplanes). (Canadian Aviation, Vol. 16, No. 7, July, 1943, p. 116.)                |
| 307  | 19910    | Germany | •••       | Wing De-Icing by Exhaust Assisted by a Combusti-<br>ble Charge (736,113). (D.F.S., Flugsport, Vol.<br>35, No. 15, 17/11/43, p. 71.)                  |
|      |          |         |           | Airport Operation.   |
| 308  | 18553    | G.B     | •••       | Experimental Soil Mechanics—Part VI. (R. Allin, Civil Engineering, Vol. 38, No. 449, November,   |
| 309  | 18961    | G.B     | , <b></b> | 1943, pp. 238-239.) Airfield Saturation (Growing Difficulty of Finding Suitable Sites). (Flight, Vol. 45, No. 1,830,                                 |
| 310  | 19113    | G.B     | •••       | 20/1/43, p. 73.)  Proposed Air Base at Southampton. (Engineering, Vol. 157, No. 4,071, 21/1/44, p. 53.)  |
| 311  | 19167    | G.B     | •••       | Portable Heatable Oil Tank for Airport. (National Petroleum News, Vol. 35, No. 47, 24/11/43, p.  |
| 312  | 19198    | G.B     | •••       | World Air Junction (Gatwick Airport Development Plan Advocated in Norman and Dawbarn's Report). (Aeronautics, Vol. 9, No. 5, Dec., 1943, pp. 40-44.) |

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| NO.  |       | REF.    |       | TITLE AND JOURNAL.   |
| 313  | 19254 | G.B.,   | •••   | Runway Materials. (Aeronautics, Vol. 9, No. 4<br>Nov., 1943.)  |
| 314  | 19387 | G.B     | •••   | Three Proposed Airports—Southampton, Portsmouth and Morecambe as Sites for Transatlantic Terminals. (Flight, Vol. 45, No. 1,831, 27/1/44, pp. 103-104.)  |
| 315  | 19397 | G.B     | •••   | Vancouver Builds an Airport. (A. F. Roberts, Commercial Aviation, Vol. 5, No. 10, October, 1943, pp. 80-84.)   |
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